

Aerodynamic Decelerator Atmospheric-entry Module (ADAM) for Mars

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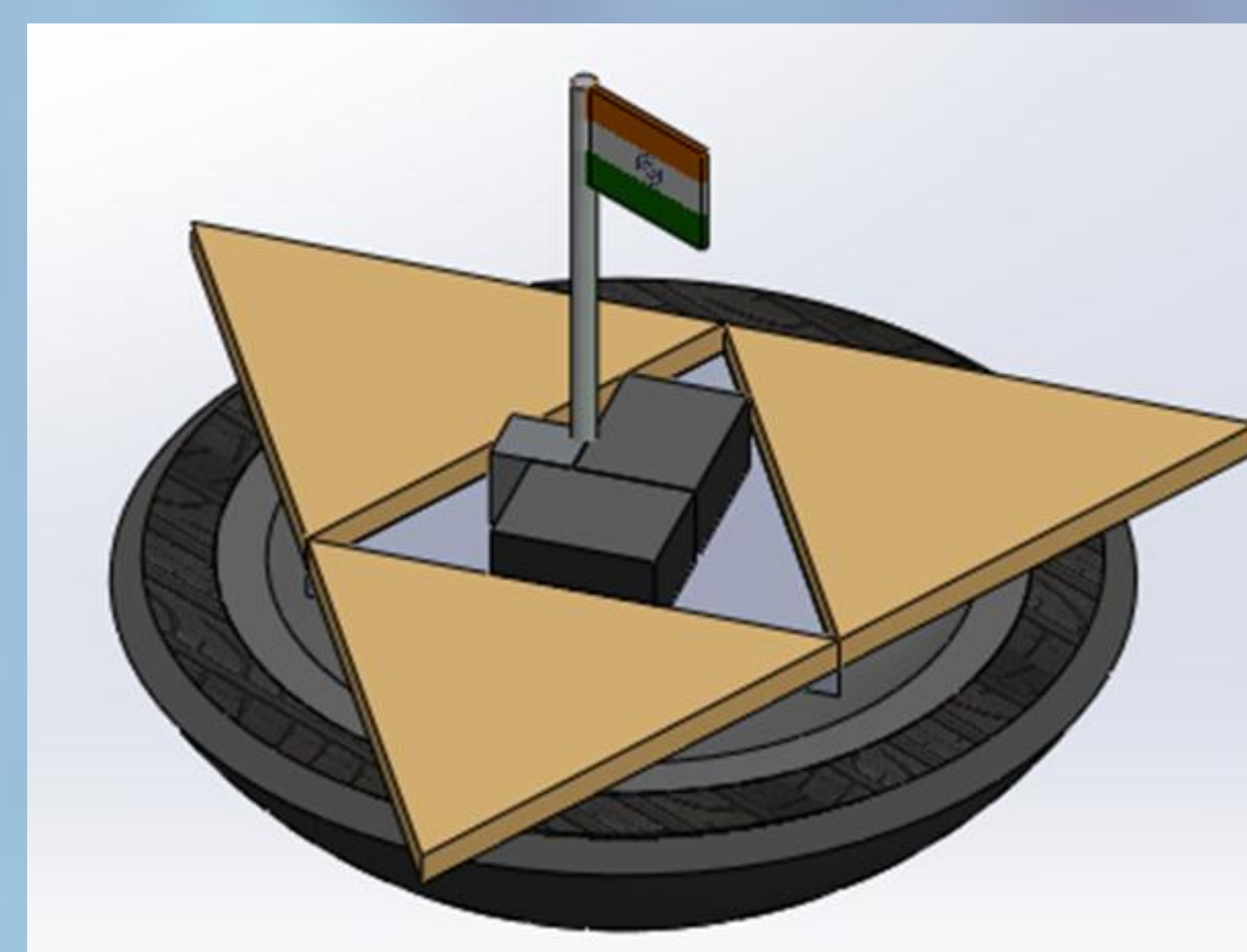
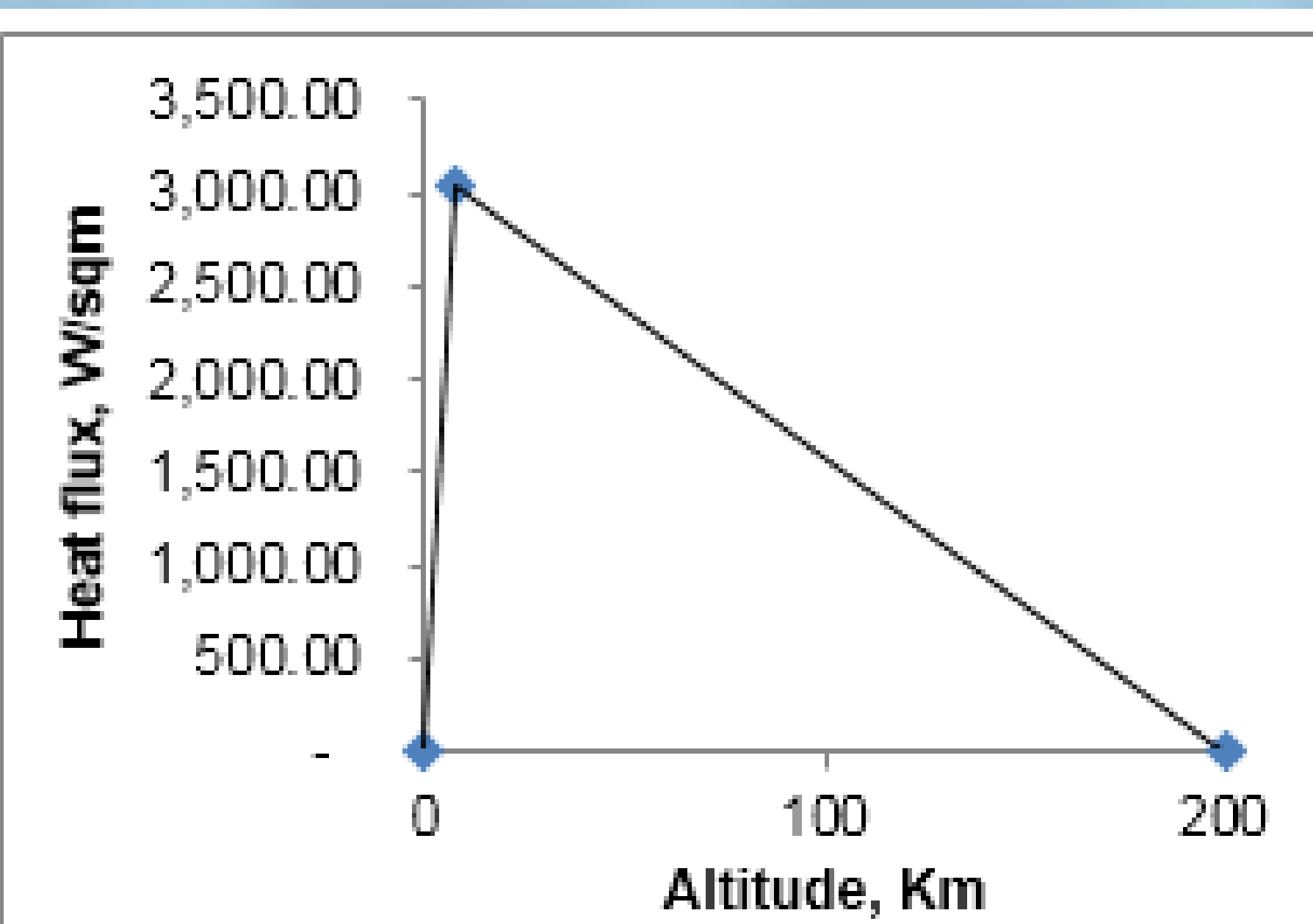
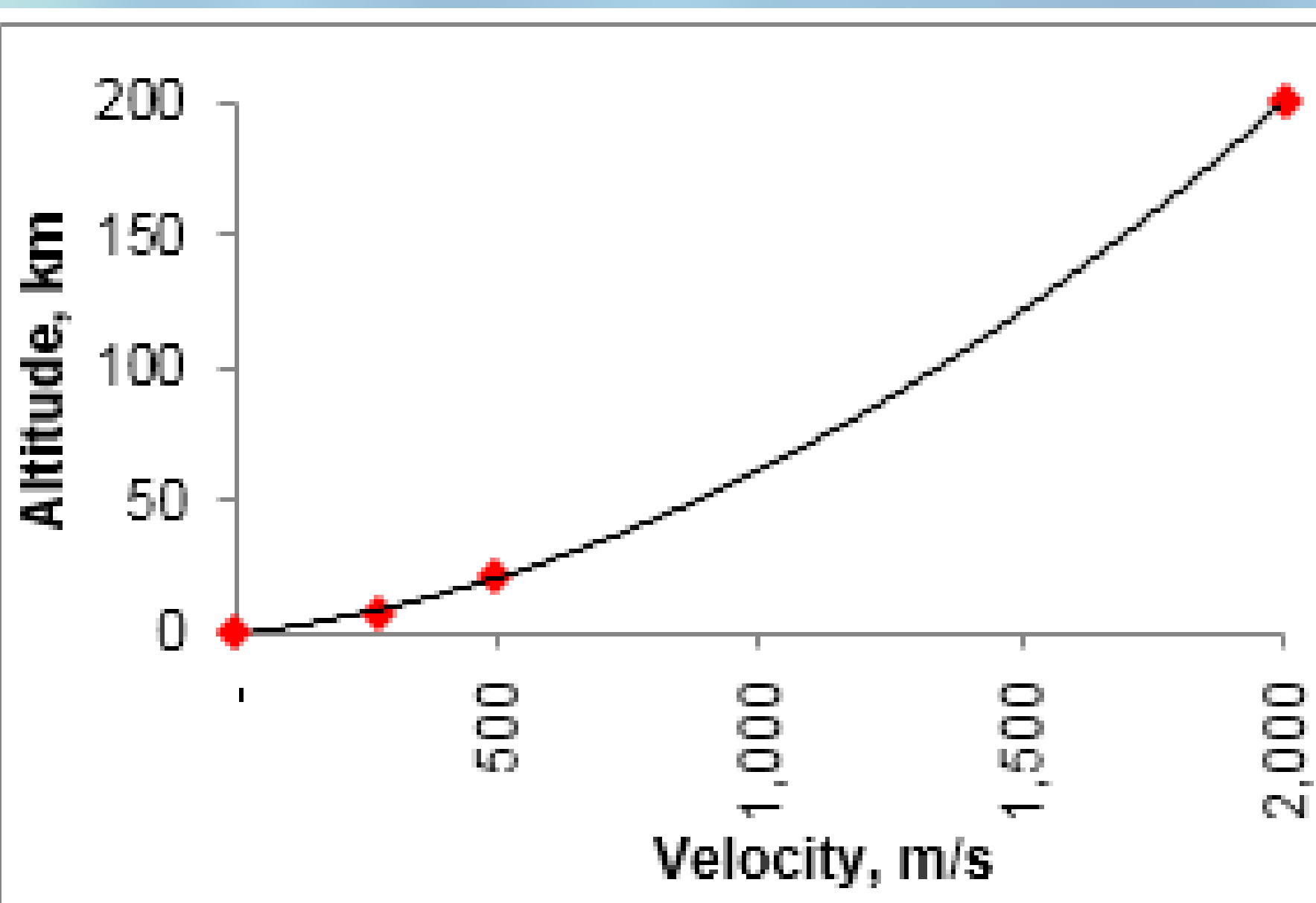
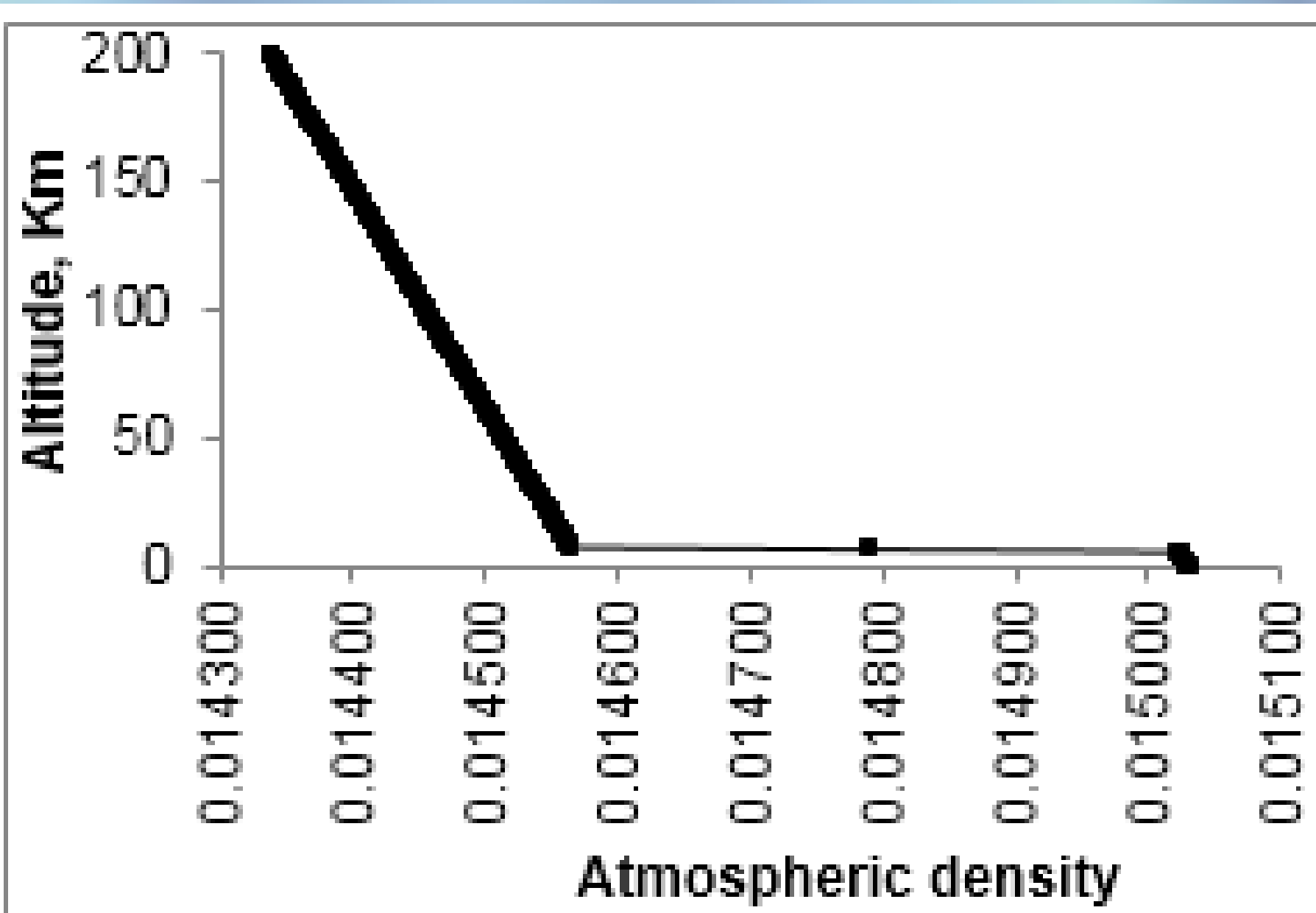
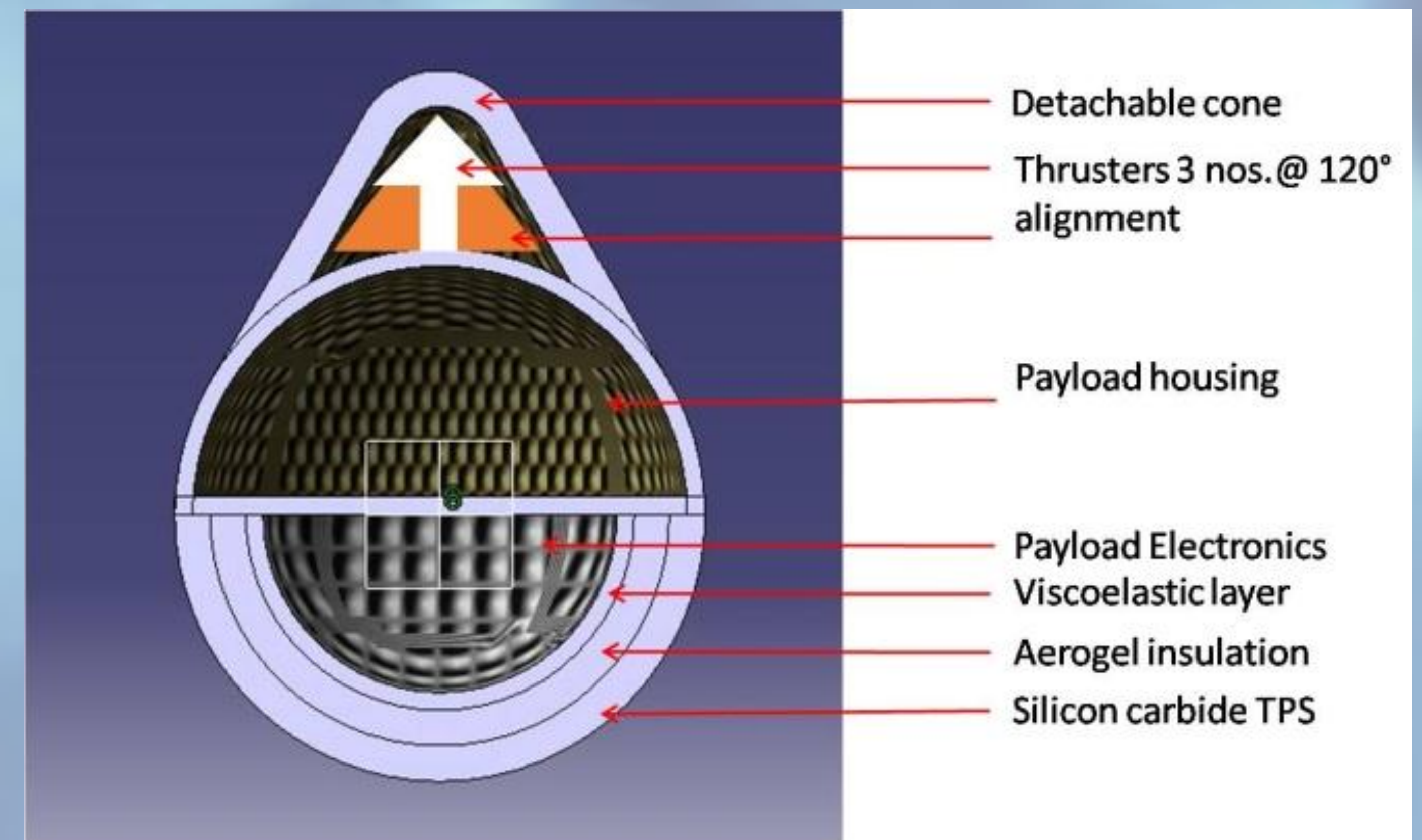
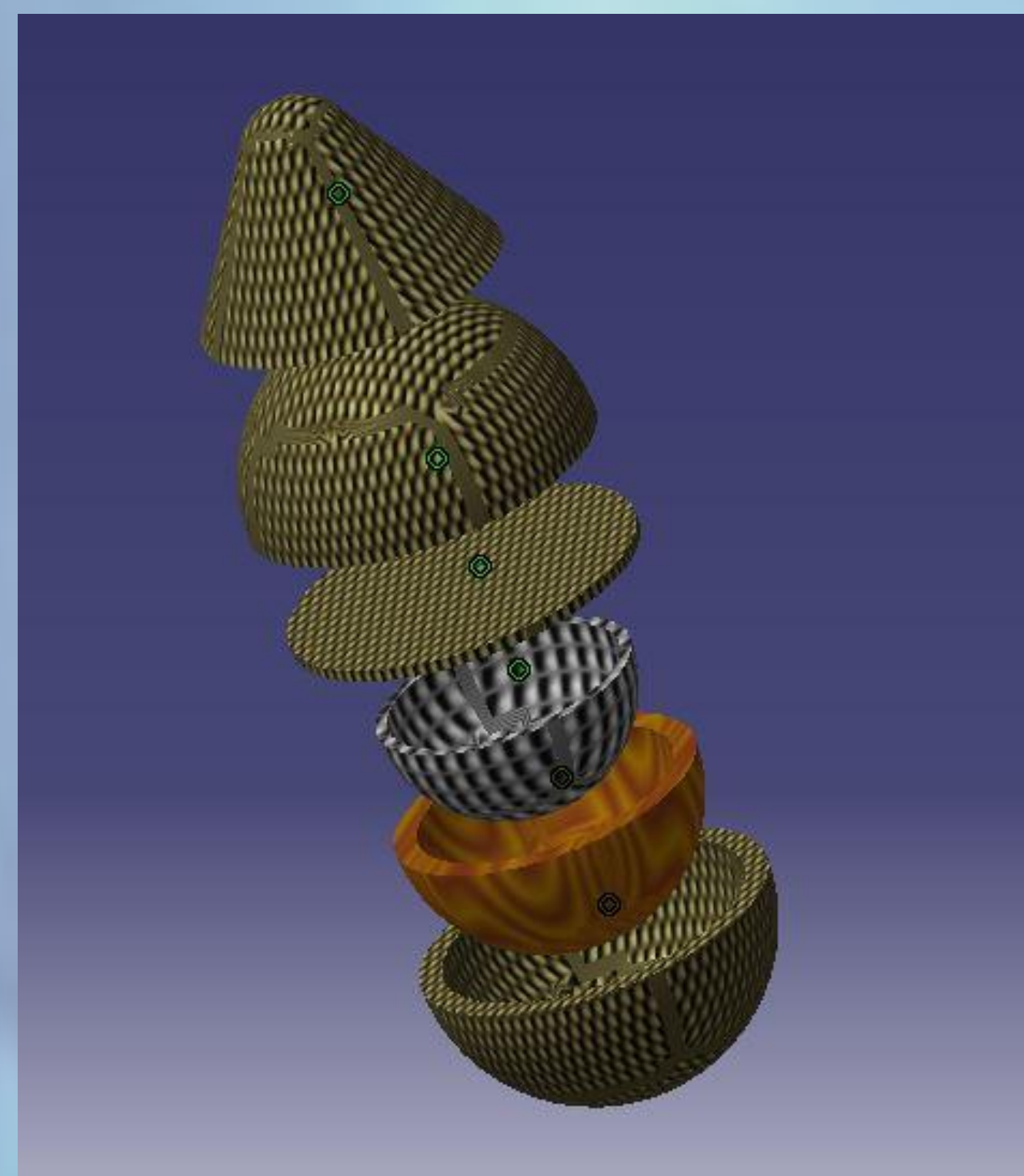
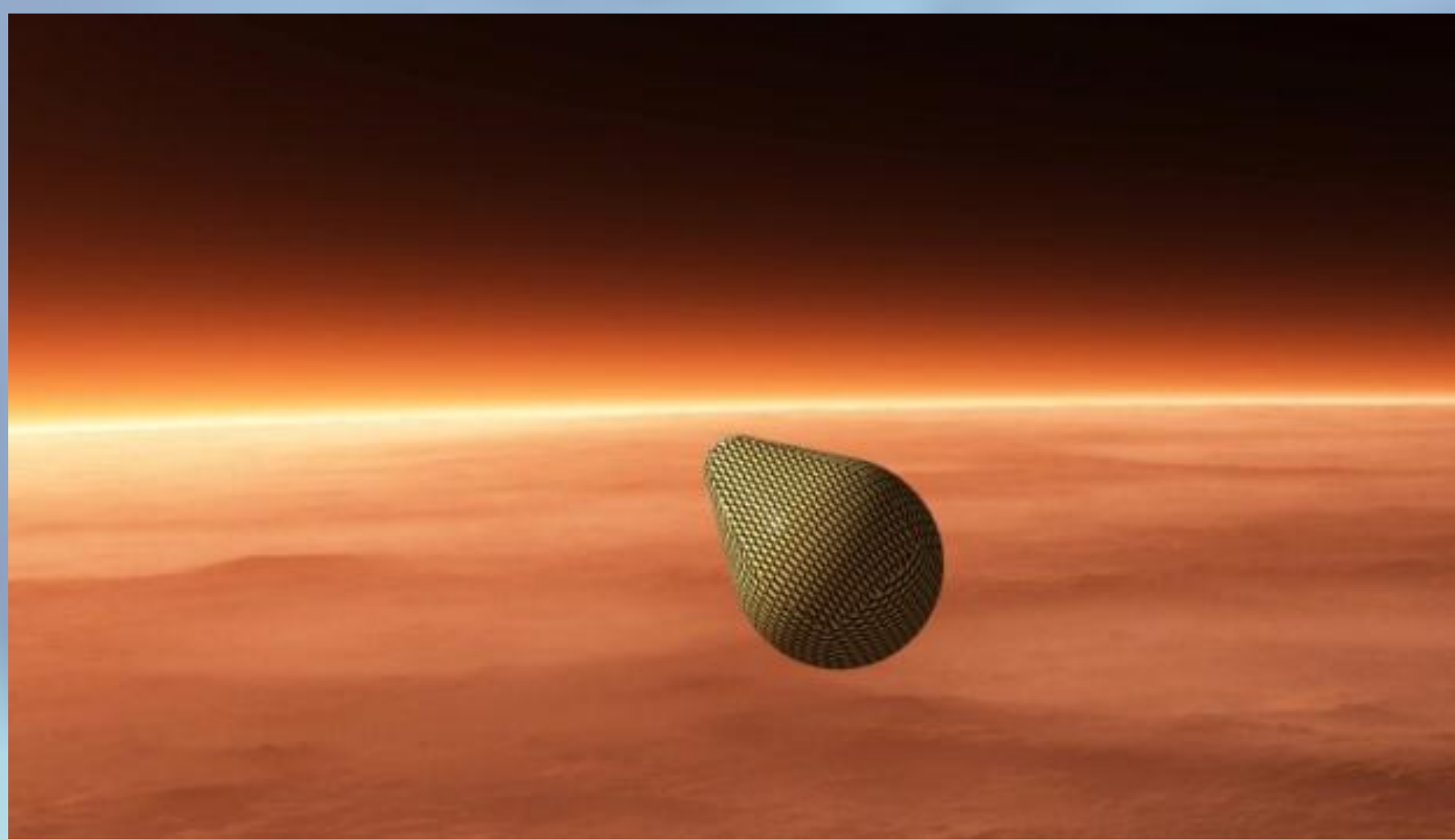
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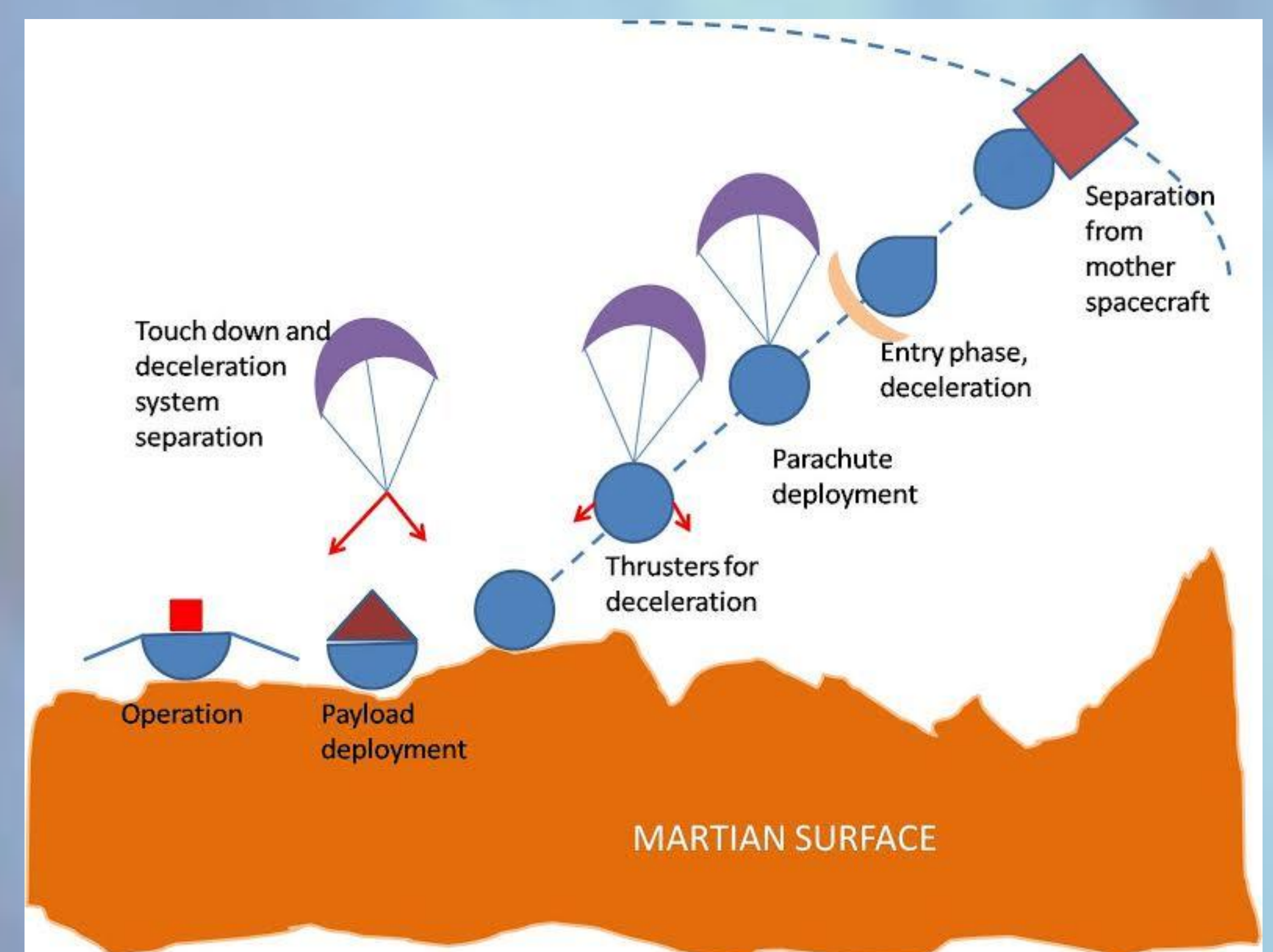
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- ADAM is an atmospheric entry module that delivers safely micro/mini probes/payloads to Martian surface, weighing from minimum 5 kg to 15 kg net weight relevant for small scale researchers and explorers.
- Conceptualized as a stretched sphere (tear drop shape during its atmospheric entry) and the rear cone separates out during the parachute deployment transforming the module to a sphere.
- This sphere drops down decelerating, aided with three thrusters to bring down the velocity to near zero value for a safer touchdown.
- Visco-elastic damping materials serves as shock absorbers to prevent impact load transfer to the payload inside.

- This paper aims primarily of...
- An atmospheric entry probe undocked from the mother spacecraft orbiting the planet
- The probe enters the Martian atmosphere decelerating itself to a near zero touchdown velocity on the surface of the planet.
- A spherical configuration is proposed for the probe having an overall weight of 15 kg with the payload inside.



- Outer aeroshell in the shape of a stretched sphere (tear drop shape)
- Multiple layer of Thermal Protection System (TPS)
- Visco-elastic shock absorbing layer
- Decelerator system consisting of parachute and thrusters



- Maximum deceleration: 11.5 m/s (3.28 'g')
- Altitude of maximum deceleration: 20.40 km from surface
- Velocity at maximum deceleration: 491.93 m/s
- Maximum dynamic pressure: 880 Pa
- Drag force at maximum deceleration: 180 N
- Altitude of maximum heating: 8.20 km from surface
- Maximum heating rate: 3033 Watt/sqm (lesser than the space capsule recovery experiment heating rate as achieved by ISRO in earth atmosphere)

- Robinson H Jennifer, Orbital Debris Impact Damage to Reusable Launch Vehicle, NASA, AL 35812 USA
- Tauber E. Michael, A Review of High Speed Convective, Heat-Transfer Computation Methods, Ames Research Center NASA Technical paper 2914, 1989
- Kazuhiko Yamada et al., Re-entry Demo Plan of Flare-type Membrane Aeroshell for Atmospheric Entry Vehicle using a Sounding Rocket, AIAA Conference, Dublin, 2011

- Concept is analogous to advent of mini-satellites when larger spacecrafts of agencies reign the space arena.
- Small scale space enthusiasts can enter planetary exploration via such small payload delivery systems